The Group of Responsables Aerodynamics (GoR AD)
An Overview of activities and Success Stories

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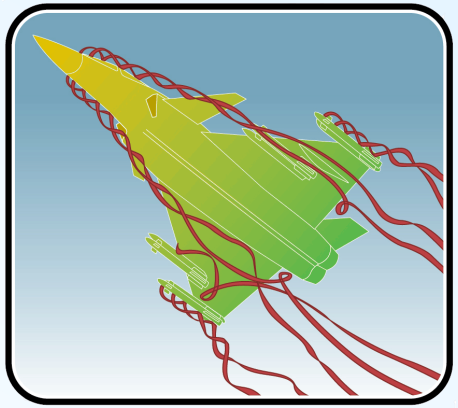
GARTEUR Story and Mission

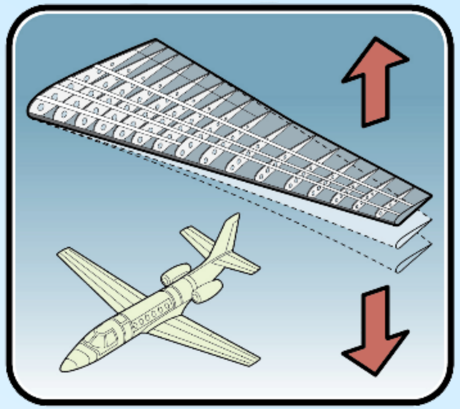


Aerodynamics

Aviation security

Flight mechanics system and integrations

Rotorcraft



Structures and materials

For each area a Group of Responsible (GoR) is established with the objective to address and monitor activities (Exploratory Group and Action Group)

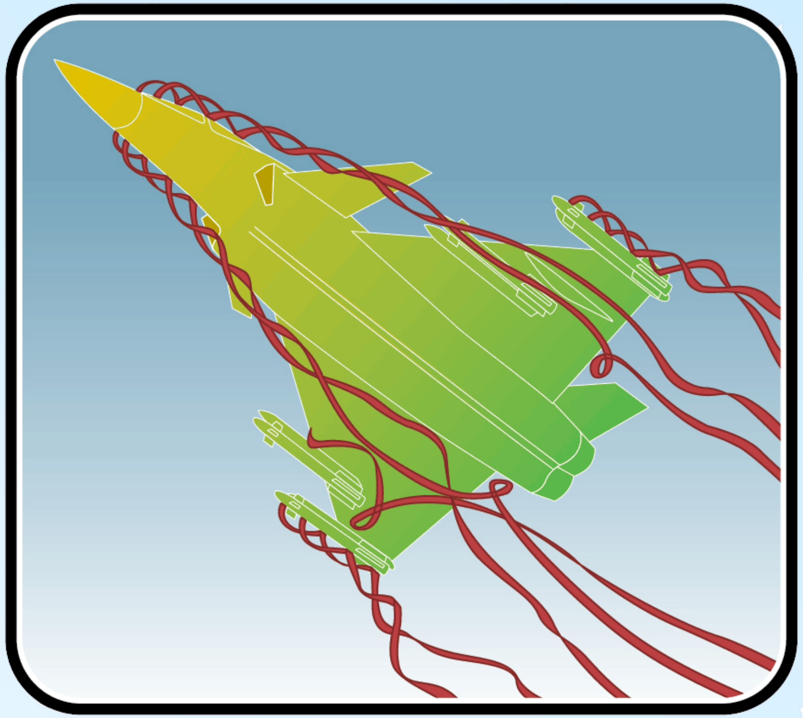
Aerodynamics GoR

The Group is active in experimental, theoretical, analytical, as well as in numerical fields of aerodynamics to support the development of methods and procedures.

Work in experimental areas is performed mainly to obtain valuable data for the validation of methods. Measurement techniques are developed and refined to increase accuracy and efficiency of experimental investigations. Numerical studies and development of CFD codes are performed to give insight in the mechanisms of basic flow phenomena.

The GoR AD initiates and organizes basic and applied aerodynamic research in the field of aeronautics.

Aerodynamics GoR



Topics:

- ✓ Aerodynamics
- ✓ Aero-thermodynamics
- ✓ Aero-acoustics
- ✓ Aero-(servo-)elasticity
- ✓ Aerodynamic shape optimization
- ✓ Aerodynamics coupled to flight mechanics
- ✓ Aircraft icing simulation
- ✓ Multidisciplinary design and analysis
- ✓ Systems and propulsion aerodynamic integration

Both experimental and computational. Both civil and military.

- ✓ GARTEUR along the time had a fundamental role in establishing cooperation among European company.

- ✓ When GARTEUR was set-up, in a period where collaborative projects funded from EU were far from arrive, GARTEUR represented a unique possibility of cooperation among European companies and research centres and represented the feed of future European funded collaborative projects.

  - ✓ The first GoR-AD action group is from 1979: AD/AG-01 wing body aerodynamics at transonic speed

  - ✓ At present time we have arrived at AD/AG-61 WMLES and Embedded LES.

  - ✓ Therefore this means that along its life GARTEUR-AD has developed 61 projects on several topics related to aerodynamics

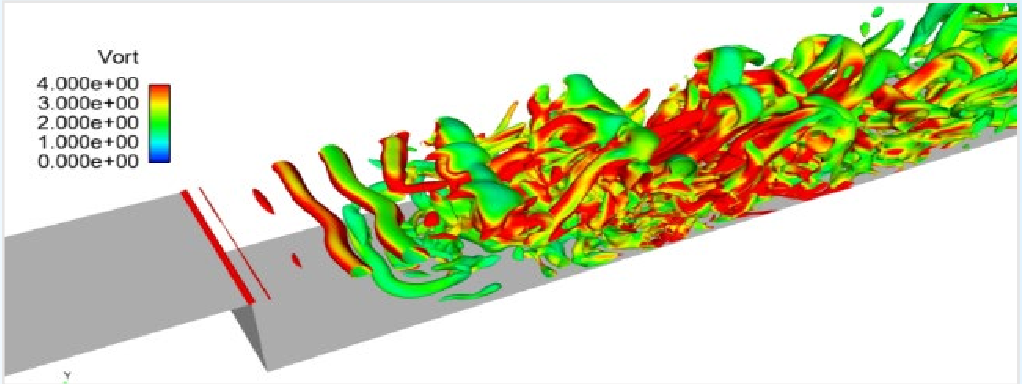
PERFORMED ACTION GROUP (1/2)

AD/AG-01 Wing-body aerodynamics at transonic speed
AD/AG-02 Two dimensional transonic testing methods
AD/AG-03 Theory/Experiment comparison for high lift airfoil (High lift phase I)
AD/AG-04 An experimental and theoretical investigation into the asymmetric vortex flows characteristics of bodies of revolution at high angles of incidence in low speed flow
AD/AG-05 Convergence study for transonic flow for 3D wings
AD/AG-06 Model Support Interference in Large Low-Speed Wind Tunnels
AD/AG-07 Experimental investigation of the turbulent shear layers on swept wing
AD/AG-08 High lift action group (High lift phase I)
AD/AG-09 Flow past missiles afterbody
AD/AG-10 Flow computation for advanced propellers
AD/AG-11 Computations of 2D Navier-Stokes Equations
AD/AG-12 Comparative Investigation of Predictive Capability of Aeroacoustics Methods of Single Rotation Propellers
AD/AG-13 High lift phase III
AD/AG-14 Transition for Airfoils and Wings
AD/AG-15 Validation of Euler codes for supersonic flow (2 parts)
AD/AG-16 Development of Software Package for Graphic Visualization of Flow Simulation
AD/AG-17 Verification of 3D Transonic Euler Methods for Complex Geometries
AD/AG-18 Adaptive wall wind tunnels
AD/AG-19 Particle Image Velocimetry
AD/AG-21 Pressure Sensitive Paint
AD/AG-22 Practical Application of LDV
AD/AG-23 Three Dimensional Turbulent Shear Layer Experiment, Phase 2
AD/AG-24 Navier-Stokes calculations of the supersonic flow about slender configurations
AD/AG-25 Computational Methods for High Lift Flows (High lift phase IV)
AD/AG-26 Navier Stokes Computations of 3D Transonic Flow for a Wing/Fuselage Configuration
AD/AG-27 Transition on airfoils and infinite swept wings with regard to nonlocal instability investigations
AD/AG-28 Transonic Wing/Body Code Validation Experiment
AD/AG-29 Three-dimensional turbulent shear layer experiment - Phase 3
AD/AG-30 CFD for Supersonic Civil Transport high-lift evaluation and configuration development
AD/AG-31 Analysis of a Supersonic Transport Configuration with and without foreplan using a Navier Stokes solver
AD/AG-32 Prediction of performance degradation due to icing for 2D configurations
AD/AG-33 Ice accretion prediction on aircraft components

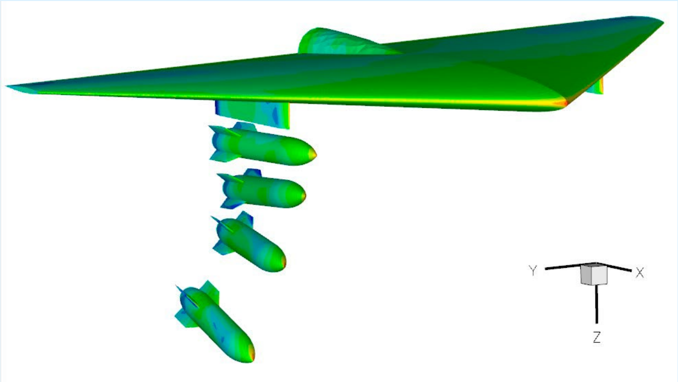
PERFORMED ACTION GROUP (2/2)

AD/AG-34 Aerodynamics of supersonic air intakes
AD/AG-35 Application of Transition Criteria in Navier Stokes Computations
AD/AG-36 3D High Lift Computations
AD/AG-37 Pressure sensitive paint, Phase II
AD/AG-38 Time Accurate Methods
AD/AG-39 Transonic Wing/Body Calculations, Phase II
AD/AG-40 Ice shapes effects on the aerodynamic performance of airfoils, Phase II
AD/AG-41 Ice Accretion Prediction, Phase II
AD/AG-42 Navier-Stokes Calculations of the Transonic Flow Over Slender Configurations
AD/AG-43 Application of CFD to High Offset Intake Diffusers
AD/AG-44 Application of Transition Criteria in NS-Computations, Phase II
AD/AG-45 Application of CFD to predict high “G” loads
AD/AG-46 Highly Integrated Subsonic Air Intakes
AD/AG-47 Coupling of CFD with Flight Mechanics
AD/AG-48 Lateral Jet Interactions at Supersonic Speeds
AD/AG-49 Scrutinizing Hybrid RANS-LES Methods for Aerodynamic Applications
AD/AG-50 Effect of Wind Tunnel Shear Layers on Aero-acoustics Measurements
AD/AG-51 Transition and Turbulence in Hypersonic flows
AD/AG-52 Surrogate Modelling in Aeronautical Design and Optimization
AD/AG-53 Receptivity and Transition
AD/AG-54 RANS-LES Interfacing Hybrid for Hybrid RANS-LES and embedded LES approaches
AD/AG-55 Countermeasures Aerodynamics
AD/AG-56 Coupled fluid dynamics and Flight Mechanics
AD/AG-57 Secondary inlets and outlets for ventilation
AD/AG-58 Supersonic air intakes
AD/AG-59 Laminar separation bubbles
AD/AG-60 Machine learning and data-driven approaches for aerodynamic optimization and uncertainty quantification
AD/AG-61 WMLES and Embedded LES

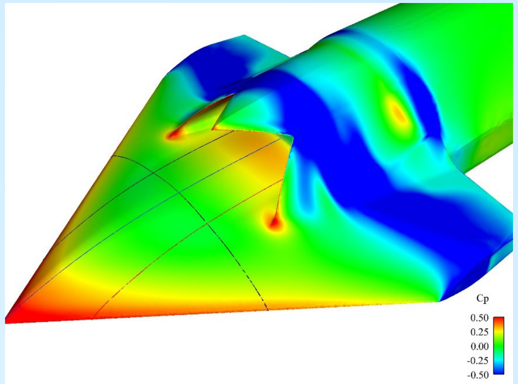
Examples of performed Action Groups



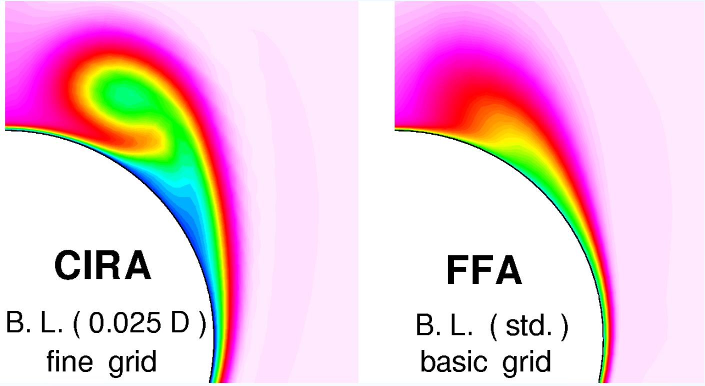
AD/AG-54 RANS-LES Interfacing Hybrid for Hybrid RANS-LES and embedded LES approaches



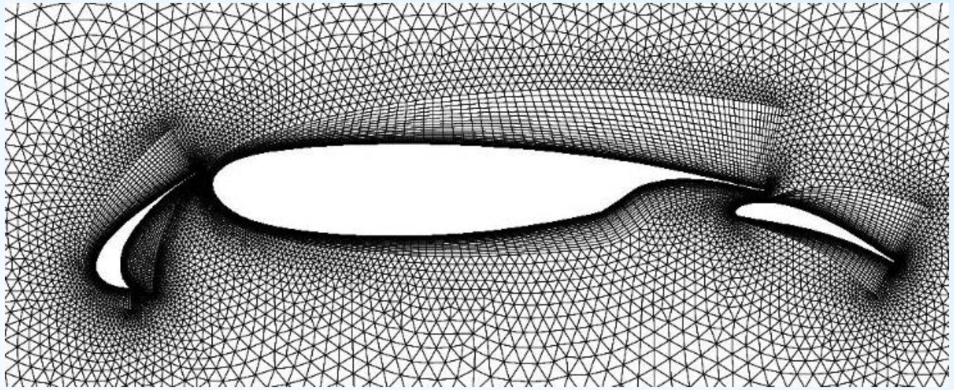
AD/AG-47 Coupling of CFD with Flight Mechanics



AD/AG-46 Highly Integrated Subsonic Air Intakes



AD/AG-24 Navier-Stokes calculations of the supersonic flow about slender configurations



AD/AG-36 3D High Lift Computations

Along this period several radical revolutions have happened in aerodynamics

- ✓ Development and availability of increasing CFD capabilities
- ✓ Design and optimization of transonic aircraft
- ✓ High lift improvements
- ✓ Studies on second generation high speed transportation
- ✓ Reduction of environmental impact
- ✓ Stealth requirements on aerodynamics design

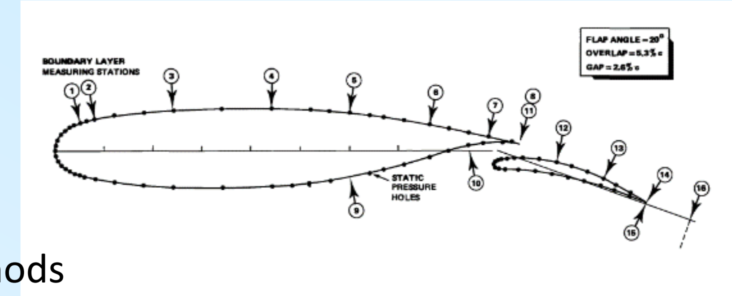
- ✓ The first one is of course the development of computer and, as consequence in parallel, the development of Computational Fluid Dynamics.

- ✓ In parallel with increase of computer resource more and more accurate tools for aerodynamics simulation have been developed.

  - ✓ Euler solver
  - ✓ RANS solver
  - ✓ Grid generation
  - ✓ LES and hybrid methods

- ✓ It is not by chance that the first action group was dedicated to numerical/experimental comparison.

The first action groups were dedicated to CFD code development and validation

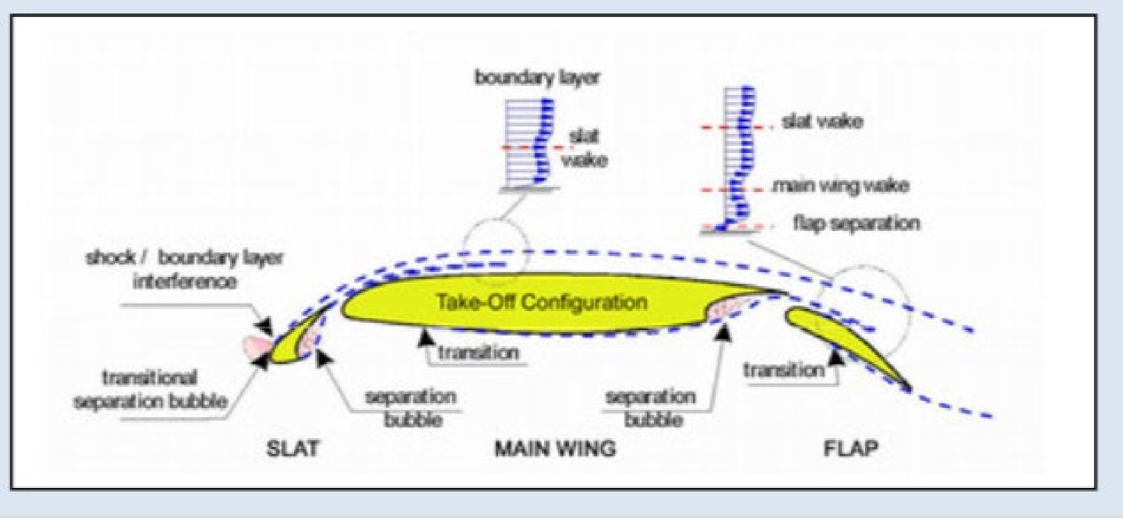


2D

- ✓ AD/AG-02 Two dimensional transonic testing methods
- ✓ AD/AG-03 Theory/Experiment comparison for high lift airfoil
- ✓ AD/AG-11 Computations of 2D Navier-Stokes Equations

3D

- ✓ AD/AG-23 Three Dimensional Turbulent Shear Layer Experiment Phase 2
- ✓ AD/AG-24 Navier-Stokes calculations of the supersonic flow about slender configurations
- ✓ AD/AG-25 Computational Methods for High Lift Flows
- ✓ AD/AG-26 Navier Stokes Computations of 3D Transonic Flow for a Wing/Fuselage Configuration



- ✓ Large efforts where dedicated to high lift:
- ✓ GARTEUR early activities on High Lift Aerodynamics were four Action Groups 1981-1998:
  - ✓ AD/AG-03 1981-1984 High lift, phase I
  - ✓ AD/AG-08 1985-1990 High lift, phase II
  - ✓ AD/AG-13 1991-1994 High lift, phase III
  - ✓ AD/AG-25 1995-1998 High lift, phase IV (maximum lift prediction)

High lift is still an open issue and more recently AD/AG-36 3D High Lift Computations was performed

In the nineties there was a new interest in supersonic transport, and therefore this interest was caught by GARTEUR that addressed the topics.

- ✓ AD/AG-30 CFD for Supersonic Civil Transport high-lift evaluation and configuration development
- ✓ AD/AG-31 Analysis of a Supersonic Transport Configuration with and without foreplan using a Navier-Stokes solver

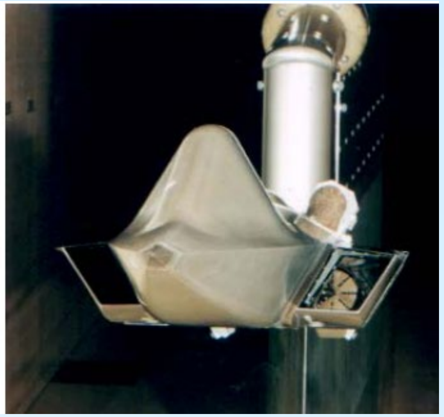
Interest on high speed transportation was replaced by interest on environmental impact, therefore, AG regarding fuel consumption reduction (laminarity) and noise have been addressed

- ✓ AD/AG-27 Transition on airfoils and infinite swept wings with regard to nonlocal instability investigations
- ✓ AD/AG-35 Application of Transition Criteria in Navier-Stokes Computations
- ✓ AD/AG-53 Receptivity and Transition
- ✓ AD/AG-12 Comparative Investigation of Predictive Capability of Aeroacoustics Methods of Single Rotation Propellers

Flight safety has also been addressed with icing topics

- ✓ AD/AG-32 Prediction of performance degradation due to icing for 2D configurations configurations
- ✓ AD/AG-33 Ice accretion prediction on aircraft components
- ✓ AD/AG-40 Ice shapes effects on the aerodynamic performance of airfoils, phase II
- ✓ AD/AG-41 Ice Accretion Prediction, Phase II

A large amount of work has been performed also for military aircraft problems



- ✓ AD/AG-09 Flow past missiles afterbody
- ✓ AD/AG-34 Aerodynamics of Supersonic Intakes
- ✓ AD/AG-46 Highly Integrated Subsonic Air Intakes
- ✓ AD/AG-48 Lateral Jet Interactions at Supersonic Speeds
- ✓ AD/AG-55 Countermeasures Aerodynamics

AG-55 Countermeasure aerodynamics

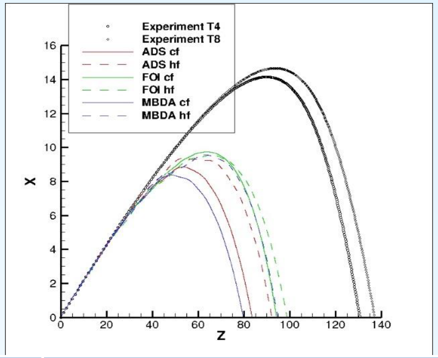
Countermeasures are used to decoy enemy tracking systems. Aerodynamics is crucial for the performance of countermeasures protecting air vehicles.

In this action group, prediction of trajectories for countermeasure objects is addressed.

Two commonly used countermeasures are chaff and flares, which are the main focus of this action group and that have been simulated by using CFD

- ✓ Chaff is radar countermeasure consisting of small pieces or threads of metal or metalized glass fibre.
- ✓ Flares are used against IR-seeking missiles. They are a few decimetres in length and can have built in propulsions systems.

a) Experimental facilities at Lacroix

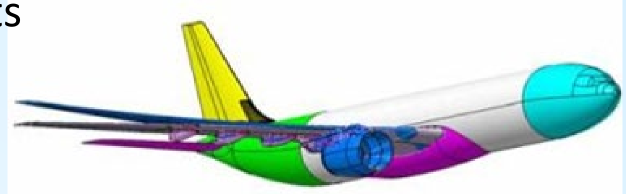


b) Comparison of measurements and simulations

AG-56 Coupled fluid dynamics and flight mechanics simulation of very flexible aircraft configurations

The goals of AD/AG-56 are twofold: firstly, this endeavour aims to enhance each partner's capabilities in aeroelastic simulations pertaining to very flexible aircraft. A second aim is to derive a common test case in terms of aircraft and manoeuvre. This will allow the various partners to benchmark their solvers and tools.

This topic poses a challenge due to various requirements inherent to such analyses:



Airbus XRF-1 model

- ✓ A flight mechanics model for flexible structures,
- ✓ CFD methods with robust grid handling technique capable of modelling a combination of large rigid body motion and large flexible motion,
- ✓ Fluid-structure interaction procedures that are capable of modelling large translations and finite rotations.

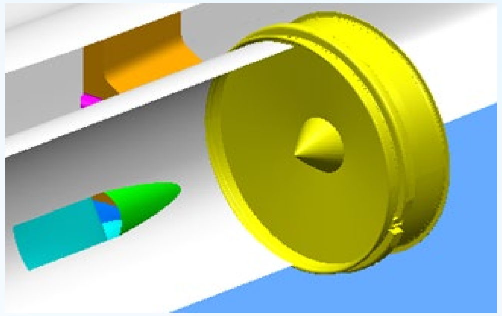
Participants: NLR (coordinator), Airbus, DLR, CIRA, ONERA,

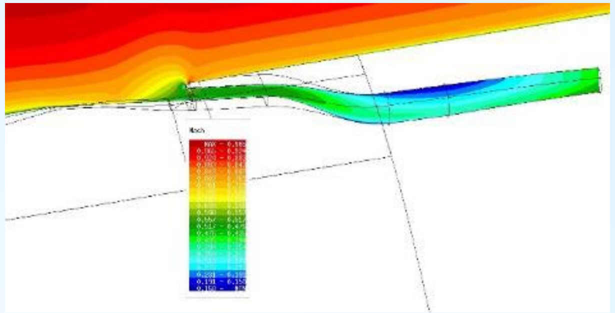
AG-57 Secondary inlets and outlets for ventilation

Analyse the efficiency of a submerged NACA type air intake for multiple flight envelopes using state-of-the-art CFD and performance evaluation methods.

Analyse the feasibility of a low-observable secondary inlet integrated in the main air intake duct of a combat aircraft and assess the impact on the intake duct flow field and on engine/intake-compatibility.

Investigate different types of secondary inlets, shapes, locations, and sizes with respect to advantages regarding radar cross section and aerodynamic performance.





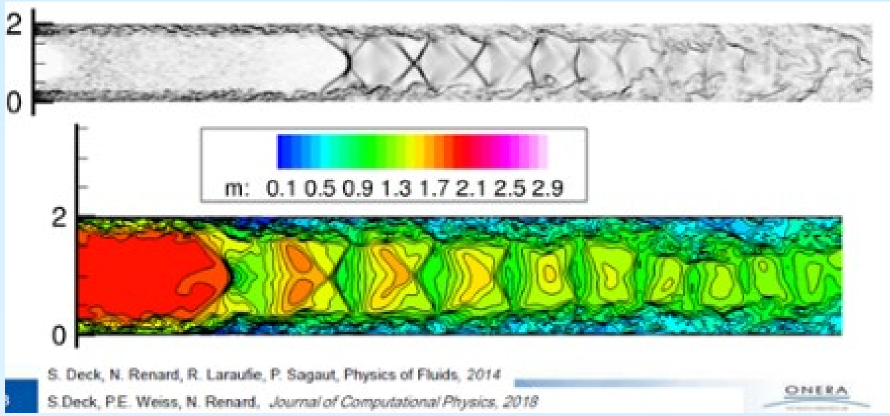
Participants: CIRA (coordinator), Airbus, DLR, NLR

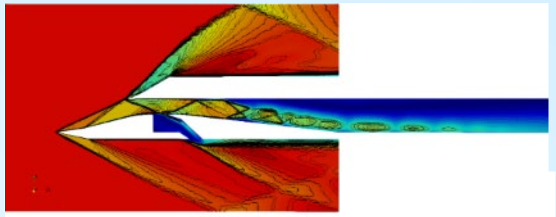
AG-58 Supersonic air intakes

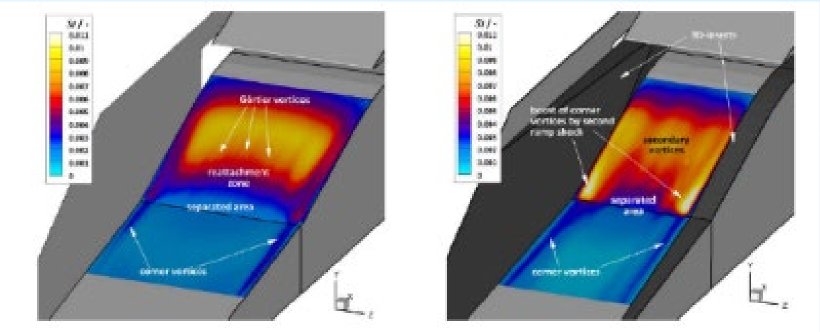
The main objective for the AG-58 is to gather a database of relevant flow features for supersonic air intakes and validate CFD codes on these specific topics. Three test cases have been identified:

- A generic supersonic diffuser flow
- A supersonic Mach 3 ramjet air intake
- An hypersonic Mach 7.5 scramjet air intake

The project is expected to yield increased understanding of turbulence modelling issues for complex internal flows in supersonic and hypersonic intakes.





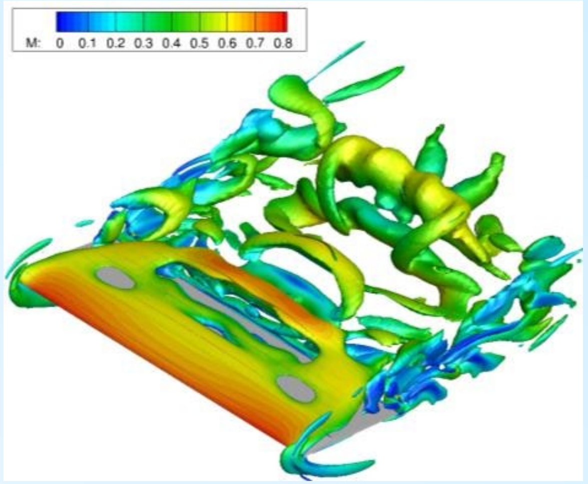


AG-59 Improving the modelling of laminar separation bubbles

The main objective is to improve the modelling of the numerical methods used in the reproduction of the laminar separation bubbles and the consequent effects on flow instability.

The main issues to be addressed are:

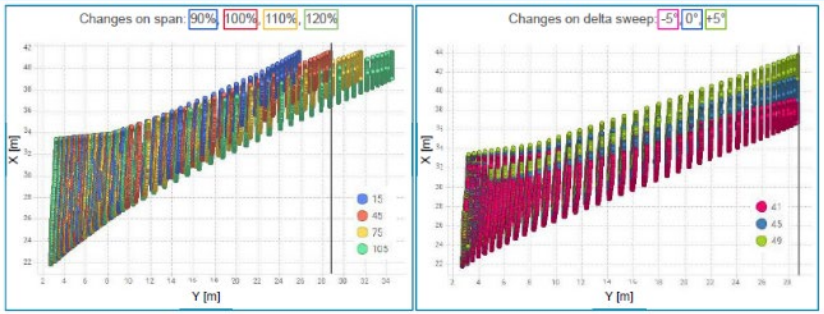
- ✓ The determination of the transition location and of the transition region,
- ✓ The enhancement of the production of the turbulent kinetic energy in the separated flow inside the recirculation region,
- ✓ Evolution of the bubble with the incidence and with turbulence level,
- ✓ Possible burst of the bubble at high incidence and consequences on the stall characteristics,
- ✓ Critical evaluation of the laminar boundary-layer instability analysis methods treatment of laminar separation bubbles.

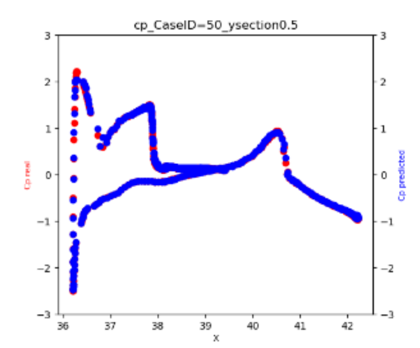


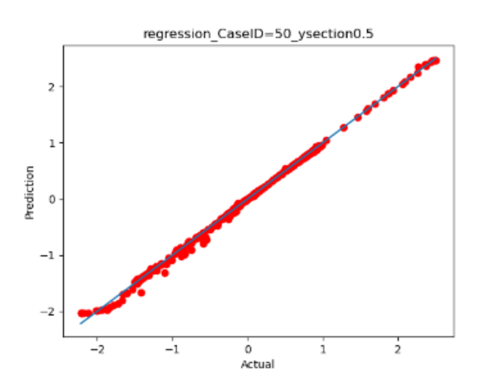
AG-60 Machine learning and data-driven approaches for aerodynamic optimization and uncertainty quantification

Fluid dynamics has traditionally dealt with massive amounts of data from experiments, field measurements, and large-scale numerical simulations.

The purpose of the AG60 is to perform an extensive comparison of surrogate models and machine learning techniques for aerodynamic analysis and prediction. The action group consists of 11 partners, including eight research establishments (CIRA, NLR, INTA, DLR, FOI, ONERA, IRT and INRIA), two industrial partners (AIRBUS-Military, AIRBUS) and one SME (OPTIMAD).







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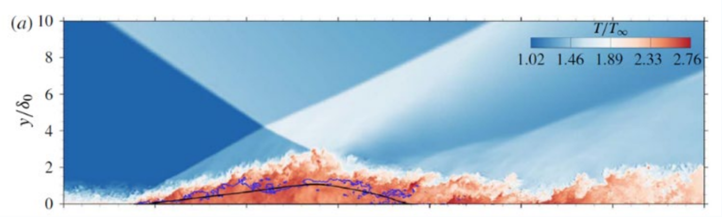
AG-61 WMLES and Embedded LES

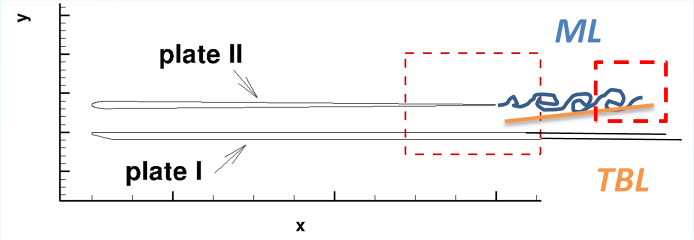
The recent needs of the aerospace industry for design and certification processes have issued increasingly difficult challenges for the CFD community. This requires simulations of increased fidelity at an affordable computational cost. An attractive way is the **Hybrid RANS/LES Methods** "HRLM" aiming to treat the regions of interest with LES (Large Eddy Simulations) whereas the rest of the flow is modelled in RANS (Reynolds Averaged Navier-Stokes) approach at a low computational cost (LES is thus embedded).

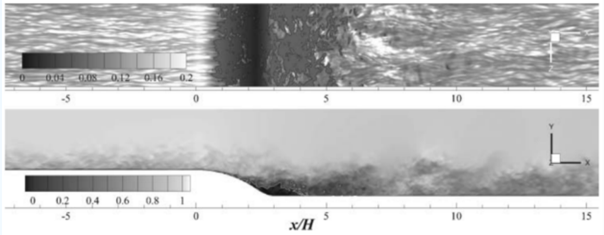
The proposed activities will be conducted by means of extensive computations of the following four test cases that will serve the modelling validation and verification. Partners: ONERA (lead), FOI, CIRA, DLR, NLR, SAAB, UNIMAN, UNISTRA

TC 1:
Mixing co-flow of wake & Boundary Layer



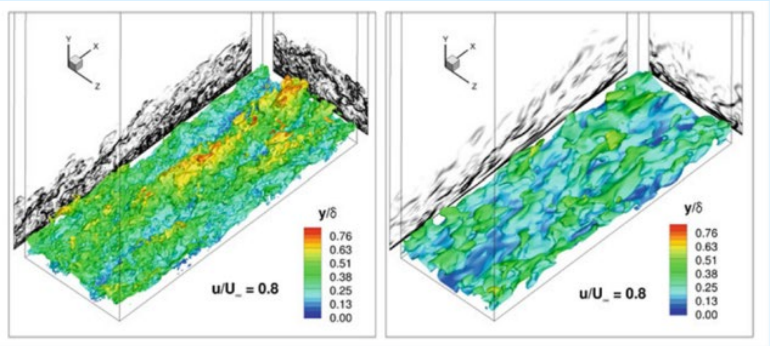
TC2: Shock Wave-Boundary Layer Interaction





TC 4: Fundamental WMLES Zero Pressure Gradient flat-plate boundary layer

TC 3:
Shallow flow separation from a smooth surface (2D-bump)



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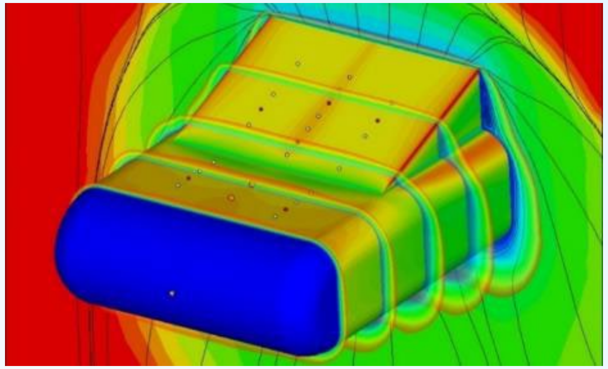
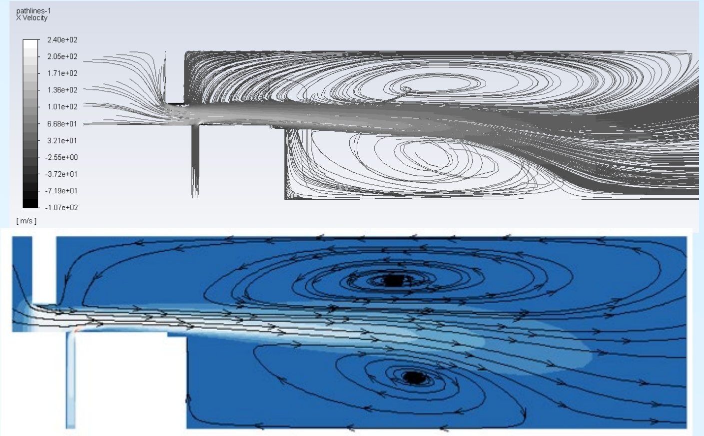
EXPLORATORY GROUP

AD/EG-79 Hypersonic *flow*
AD/EG-80 Morphing technologies
AD/EG-81 Virtual certification
AD/EG-82 Corner flows
AD/EG-83 Hydrogen combustion



EG-79 Shock wave boundary layer interaction



EG-83 hydrogen combustion